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***In the Claims***

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This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims

1. (Currently amended) A process for removing a thermal barrier ceramic coating from a metallic substrate surface of a component comprising: directing an air jet at the thermal barrier coating on the substrate surface of the component, the jet containing a non-abrasive particulate media and emitting the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient for the media to damage the substrate surface but said low pressure is sufficient for the media to remove the thermal barrier ceramic coating from the substrate surface.
2. (Previously presented) The process of claim 1 wherein the pressure of the air jet is from about 20 to 100 PSIG.
3. (Previously presented) The process of claim 2 wherein the media has a substantially spherical shape.
4. (Previously presented) The process of claim 3 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
5. (Previously presented) The process of claim 4 wherein the media is glass beads.
6. (Previously presented) The process of claim 1 wherein the component is a turbine engine component.
7. (Previously presented) The process of claim 6 wherein the turbine engine component is a combustion chamber.
8. (Currently amended) A process for removing a thermal barrier ceramic coating from a cooling hole of a metallic turbine engine component comprising: directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and emitting the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient for the media to damage a metallic surface of the cooling hole but said low pressure is sufficient for the media to remove the thermal barrier ceramic coating from the metallic surface of the cooling hole.
9. (Previously presented) The process of claim 8 wherein the pressure of the air jet is from about 20

to 100 PSIG.

10. (Previously presented) The process of claim 9 wherein the media has a substantially spherical shape.
11. (Previously presented) The process of claim 10 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
12. (Previously presented) The process of claim 11 wherein the media is glass beads.
13. (Previously presented) The process of claim 12 wherein the turbine engine component is a combustion chamber.
14. (Previously presented) The process of claim 8 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
15. (Previously presented) The process of claim 9 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
16. (Previously presented) The process of claim 8 wherein the air jet rounds the metallic edges of the cooling hole.
17. (Previously presented) The process of claim 8 wherein the cooling holes are drilled into the turbine component using a laser drilling process.
18. (Currently amended) A process for forming cooling holes on a thermal barrier coated turbine engine component comprising: drilling cooling holes into the component; coating a surface of the component containing the cooling holes with a thermal barrier ceramic coating; and directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and emitting the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient for the media to damage a metallic surface of the cooling hole but said low pressure is sufficient for the media to remove the thermal barrier coating from the metallic surface of the cooling hole.
19. (Previously presented) The process of claim 18 wherein the pressure of the air jet is from about 20 to 100 PSIG.
20. (Previously presented) The process of claim 19 wherein the media has a substantially spherical

shape.

21. (Previously presented) The process of claim 20 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
22. (Previously presented) The process of claim 21 wherein the media is glass beads.
23. (Previously presented) The process of claim 22 wherein the turbine engine component is a combustion chamber.
24. (Previously presented) The process of claim 16 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
25. (Previously presented) The process of claim 18 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
26. (Previously presented) The process of claim 18 wherein the air jet rounds the metallic edges of the cooling hole.
27. (Previously presented) The process of claim 18 wherein the cooling holes are drilled through the turbine component using a laser drilling process.